

IN THE CLAIMS

1. (withdrawn) A method for assembling a gas turbine engine, said method comprising:

coupling a combustor including a dome assembly and a combustor liner that extends downstream from the dome assembly to a combustor casing that is positioned radially outwardly from the combustor;

coupling a ring support that includes a first radial flange, a second radial flange, and a plurality of beams that extend therebetween to the combustor casing; and

coupling a primer nozzle including an injection tip to the combustor such that the primer nozzle extends axially through the dome assembly such that fuel may be discharged from the primer nozzle into the combustor during engine start-up operating conditions.

2. (withdrawn) A method in accordance with Claim 1 wherein coupling a primer nozzle including an injection tip to the combustor further comprises coupling a primer nozzle to the combustor such that fuel is discharged axially from the primer nozzle into the combustor in a direction that is substantially parallel to a centerline axis extending through the combustor.

3. (withdrawn) A method in accordance with Claim 1 wherein coupling a primer nozzle including an injection tip to the combustor further comprises coupling a primer nozzle to the combustor such that the primer nozzle extends through the ring support and includes a shroud that extends circumferentially around the primer nozzle injection tip.

4. (withdrawn) A method in accordance with Claim 1 wherein coupling a primer nozzle including an injection tip to the combustor further comprises coupling an air source to the primer nozzle such that cooling air supplied to the primer nozzle injection tip is metered by a plurality of openings extending through a shroud extending circumferentially around the primer nozzle injection tip.

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